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Article

A Study on Thermal Buckling and Mode Jumping of Metallic and Composite Plates

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Abstract: Composite plates in post-buckling regime can experience mode jumping in their buckling shape, suddenly increasing the number of half-waves. This phenomenon can be advantageous, because the shape change could be used for local morphing or structural adaptability in future aerospace structures. A study of this phenomenon under heating is here presented, combining numerical and experimental techniques. At first, a set of parametric analysis was conducted to identify composite panels that present a mode jump when heated. Three plates were selected, one in aluminum alloy 2024T3, and two in AS4/8552 composite material, with layup $[30/-30/5/-5]_s$ and $[35/-35/10/-10]_s$. The plates were tested in a new test setup for thermal buckling based on low thermal expansion fixtures. The mode jumping was successfully obtained experimentally for both composite plates. Numerical simulations predicted the general trends for all plates, and the mode jumps for the composite plates.

Keywords: post-buckling; thermal testing; Instability; experimental buckling; shape change; aeronautical morphing application



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1. Introduction

Thin-walled composite structures are present in a wide range of aerospace applications, where they are required to operate under extreme conditions. Good examples are high-speed aircraft, which must cope with combinations of thermal and mechanical loads, and satellites, which endure large cyclic temperature oscillations. Such loading conditions make these structures susceptible to buckling. This phenomenon has been largely considered as an undesirable event, due to the significant impact it has over the stiffness and the ability of a structure to carry load. The use of post-buckled states in structural design has been mainly limited, if used at all, as a contribution to increase the mass efficiency of a vehicle. In this respect, plate elements are perfect candidates for this use due to their well-known stable nonlinear behavior. Composite materials add versatility and expand the design space for the structural designer, because plate buckling and post-buckling behavior can be tailored by varying the laminate stacking orientation [1,2]. Also, their elastic behavior allows the structure to return unscathed to the original state once the load decreases.

New research tendencies are observing buckling from a different standpoint: buckled states are no longer 'forbidden', but rather an additional structural state that has properties on its own and can offer additional functionalities [3]. A new research direction uses buckling as a mechanism for shape adaptability or morphing: the load triggers a shape change without the need of any additional actuator. For instance, several authors explore the shape change through mechanically induced buckling: Vos et al. studied control actuation through post-buckled precompressed structural elements [4], and Runkel et al. [5] explored the change in torsional stiffness in a wingbox allowing buckling in one of its plate side elements. Changing the shape through heat has also been investigated: Eckstein et al. investigated the multi-stability in shell structures using temperature increments [6], and the use of thermal gradients in hybrid laminates [7].

In some situations buckled plates may experience a sudden change in buckling shape, also known as a mode jump. The topic of mode jumping or mode change has been approached in limited occasions: While the contributions to the field of simulation of composite materials are numerous, refs [8–10], the amount of papers tackling the subject are more scarce [11]. The phenomenon has been thoroughly investigated numerically and experimentally by Falzon and Hitchings [12] and Falzon and Cerini [13], who proved the suitability of nonlinear dynamic explicit procedures for the prediction of this phenomenon in uniaxially compressed composite panels, and proved the large energy release after a mode jump. In a similar fashion, even though significant contributions have been made in the experimentation of buckling in the last decade (Arbelo et al. [14,15], Bisagni et al. [16,17], or Labans et al. [18]), to the best knowledge of the authors there are no experimental studies in the field of mode jumping of heated composite plates. There is, however, an experimental study done by Ehrhardt [19] in heated metallic plates, who showed that alternative equilibrium paths were reached when an external perturbation was present.

In order to demonstrate the feasibility of mode jumping under heating as a means to obtain shape change, experimental studies become an essential milestone. A large part of available bibliography on tests in thermal buckling of plates and shells refer to high-speed applications. This is the case of Percy and Fields [20] and Thompson and Richards [21], who tested two identical hat stiffened flat plates made in titanium and titanium matrix composite under diverse combinations of heating and mechanical loads, contributing in this way to new design concepts in heated structures; Rakow and Waas [22] tested metallic foam sandwich plates using an oven, taking advantage of the differences in thermal expansion between fixture and specimen; and other authors such as Murphy et al. [23], Amabili et al. [24] or Thornton [25] performed experimental studies on buckling of heated metallic plates using diverse approaches and heating source types. Most of these tests address thermal buckling on metallic materials, and only just a few tests of carbon composite plates and shells are available in literature: Breivik and Hyer [26] tested curved composite panels under both mechanical and thermal loads, showing that temperature gradients found had a relatively small effect over the considered specimens, Fields et al. [27] performed a thermostructural tests on a titanium matrix composite panel presenting a novel experimental setup for combined loading, and Wu and Gürdal [28] performed heating tests on carbon composite, fiber steered panels, as a first step towards the application of fiber tailoring in heated aerospace structures. More recently, Xu et al. [29] tested the thermal buckling of a flat panel of carbon fiber-reinforced, silicon carbide ceramic matrix composite.

The goal of this research is to study the feasibility of a new morphing possibility in which shape changes are induced into structural composite plates by heating and without the contribution of any external actuator. This phenomenon can take advantage of post-buckling regime and the mode jumps at higher load levels. This could offer advantages and complement existent or future designs of high-speed aircraft, or also deployable structures in which low weight and multi-functionality are desirable.

A methodology for selecting laminates that present mode jumping in their nonlinear post-buckling regime is described and applied. As a result, two composite plates are selected and subsequently tested in a novel experimental setup. Prior to that, a preliminary test of an aluminum alloy plate is also performed. Results are presented in the form of deflection curves and out-of-plane deformation plots.

In particular in Section 2 the mode jumping phenomenon is explained in detail; while in Section 3 a methodology for selecting laminates that present mode jumping in their nonlinear post-buckling regime is described and applied, leading to the selection of two AS4/8552 composite plates for experimentation. In Section 4 a novel experimental setup for thermal buckling of plates is presented; and in Section 5 the results of a preliminary test on a plate made in 2024T3 aluminum alloy and of the tests on both composite plates are reported in the form of deflection curves and out-of-plane deformation plots. Finally, all findings of this investigation are summarized in the conclusions in Section 6.

2. Mode Jumping on Heated Composite Plates

Mode jumping, also called mode switch or mode change, can be defined as a sudden variation in the post-buckled shape of a loaded panel, characterized by a fast switch from one buckling mode into another one [15]. In this study, mode changes caused by temperature increments are investigated. The studied phenomenon can be qualitatively described with the help of a generic flat plate, shown in Figure 1. The plate has length a , width b and thickness h , and is located in a Cartesian system XYZ , with XY coordinates in plane with the mid-plane of the plate. In-plane size variations along X , Y axis are constrained and rotations around plate edges are also constrained. The plate is subjected to a homogeneous temperature increment ΔT that is applied in a quasi-static manner.

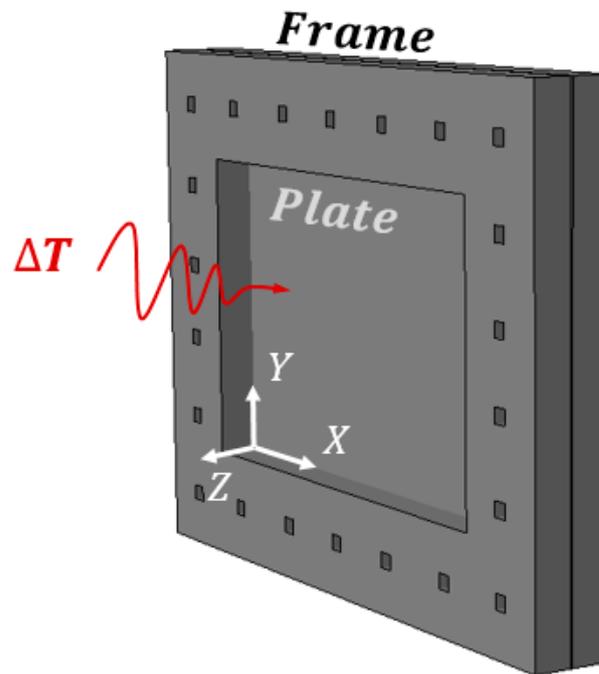


Figure 1. Plate geometry, boundary conditions and heating load.

The response of the plate can be represented in a graph as reported in Figure 2, with temperature variation ΔT versus out-of-plane position Z of the point at the middle of the plate. Initially, the plate is considered to be at a stress-free state. Applying temperature increments ΔT , in-plane stresses arise as a consequence of restrained thermal expansion, and the plate starts to follow the equilibrium path reported as a bold black line. The plate, however, remains flat up until when its equilibrium path finds a bifurcation point. Temperature at this location is represented as ΔT_{b1} , and is known as the first bifurcation temperature or also buckling temperature. For temperatures beyond this value the plate starts deforming out of plane, typically in a half-wave buckling shape. The plate is now in post-buckling state and therefore under non-linear behavior. If temperature keeps increasing, plates with some particular stacking orientation may experience a sudden change into a different, stable buckling shape. This event is commonly known as mode jumping. The temperature indicated as ΔT_{b2} in the graph is known as mode jumping temperature, and the plate changes usually from a half-wave buckling configuration to two half-waves.

The described equilibrium path corresponds to a plate without imperfections. This is, however, an ideal case and in reality geometrical imperfections are always present. Due to this, the post-buckling behavior may differ from the ideal case. An example of a plate with initial geometric imperfections is also illustrated in Figure 2 reported as a blue dashed line. Depending on imperfection shape and amplitude, equilibrium path will vary, altering not only the nonlinear deflection path but also the mode jumping temperature

ΔT_{b2} . Additionally, other factors such as size variation due to expansion of the boundary conditions may have a significant impact in the nonlinear behaviour of the plate.

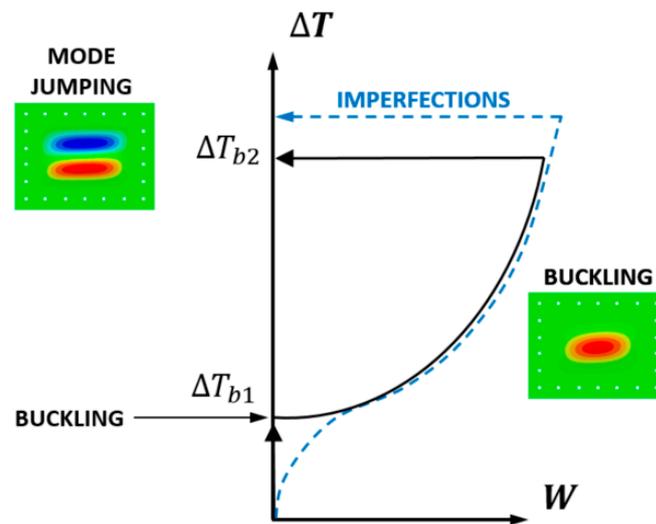


Figure 2. Thermal equilibrium path of a post-buckled composite plate.

3. Analysis to Identify Composite Plates Presenting a Mode Jump under Heating

To select configurations of composite plates that achieve a mode jump under heating, three steps were conducted. First, a set of design requirements was established. Second, the design space was inspected for the existence of plates showing mode jumping through a parametric analysis in which both linear buckling and post-buckling behavior were examined. Third, the stacking orientations were improved by fine-tuning the layups.

In the first step, the design space was established by imposing geometry and layup constraints. A fixed plate size of 200 mm × 300 mm compatible with the available oven was imposed. The plate was also required to have symmetric and balanced stacking orientation with a total number of 8 plies. The material AS4/8552 [30], with properties reported in Table 1, was used for all analyses. This is a common carbon composite material which is not suitable for high temperatures, so a temperature limit of $\Delta T = 120$ °C was imposed for the tests. As a consequence mode jumping temperature ΔT_{b2} can never exceed this value.

Table 1. Material properties.

	E_{11} (GPa)	E_{22} (GPa)	G_{12} (MPa)	ν_{12}	α_1 ($\frac{\mu\text{m}}{\text{m}^\circ\text{C}}$)	α_2 ($\frac{\mu\text{m}}{\text{m}^\circ\text{C}}$)	t_{ply} (mm)	ρ ($\frac{\text{kg}}{\text{m}^3}$)
AS4/8552	135	9.68	5.6	0.30	0.28	28	0.181	1.58
Invar 36		140		0.33		1.5		8.05
2024T3		72.3		0.33		24		2.73

Assuming that in-plane plate size variations are perfectly constrained, the thermal buckling of a given composite laminated plate is governed by the laminate coefficients of thermal expansion (CTE) as well by the bending and membrane stiffness. Taking a generic angle-ply laminate $[\theta / -\theta]_{2s}$, and using laminate theory, the variation of the bending stiffness components D_{11} and D_{22} , and the CTE α_x and α_y can be plotted versus the generic angle θ . These are reported in Figure 3a,b, respectively. It is possible to note that for small values of angle θ , the bending stiffness along X axis D_{11} is large, with its maximum at zero degrees, while CTE along X direction α_x is very low. Conversely, D_{11} is minimized at 90°, while α_x is maximized. Thus, bending stiffness and thermal expansion show opposite trends, and a balance between these two factors must be achieved. In reality, boundary conditions will expand, stretching the plate and changing the buckling temperature. Due

to this, the effect of boundary conditions was considered for the next steps in the laminate selection process taking into account the plate frame in the analyses.

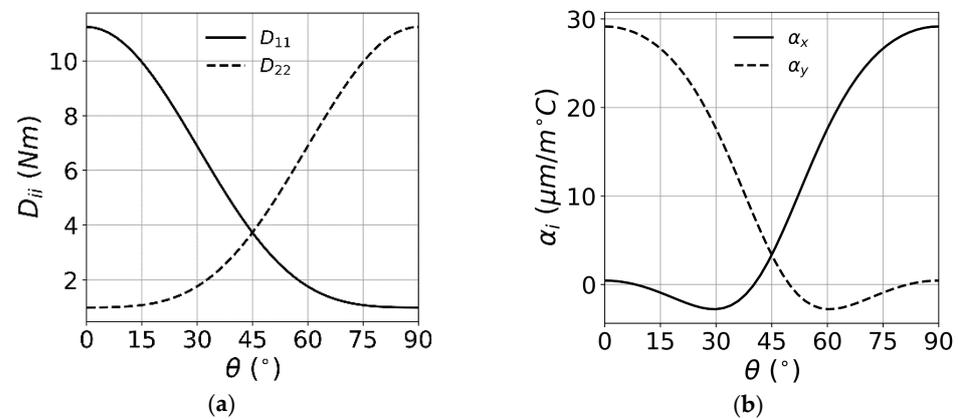


Figure 3. Properties of composite plates: (a) Bending stiffness for $[\theta / -\theta]_{2s}$ laminates; (b) CTE's for $[\theta / -\theta]_{2s}$ laminates.

In the second step, the defined design space was examined for plates presenting mode jumping in their post-buckling range. At the beginning, a generic symmetric and balanced $[\theta / -\theta]_{2s}$ stacking orientation was chosen. The laminates were studied using two finite element (FE) analyses: the buckling temperatures were calculated through a linear buckling analysis, while the simulations for plate nonlinear behavior were performed using a dynamic explicit procedure. All simulations were performed with the FE model reported in Figure 4 and using the software Abaqus (Dassault Systemes, Paris, France).

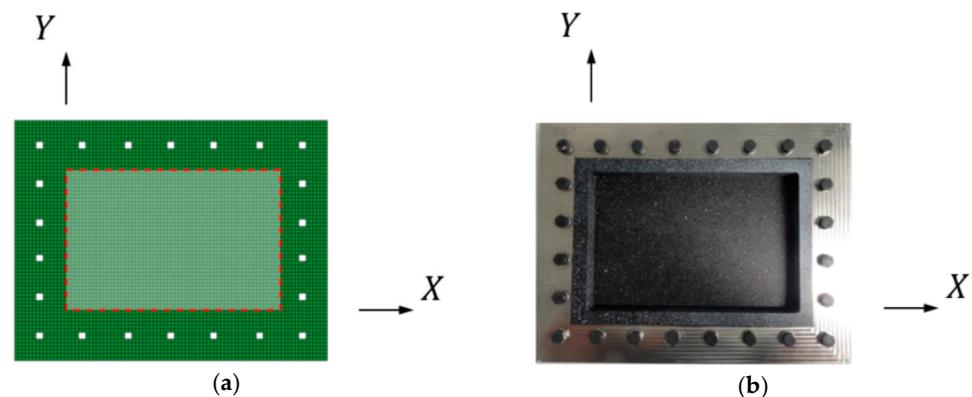


Figure 4. Plate with frame: (a) FE model; (b) Specimen.

The models present dimensions of $440 \text{ mm} \times 340 \text{ mm}$. Boundary conditions were introduced considering the surrounding structural frame to which the plate is attached. The frame overlaps with the outer area of the plate and has a cross section with width of 70 mm, so that the plate free area is $300 \text{ mm} \times 200 \text{ mm}$. The frame was made from Invar 36 material, which material properties are reported in Table 1. Plate and frame were meshed using S4R element. As the overlapping meshed areas corresponding to the plate and the frame have equal element size and nodes at identical positions, they were connected by means of a simple nodal merging operation. The influence of element size over the linear buckling temperature is reported in Figure 5, where both error and normalized computational time are plotted versus element size.

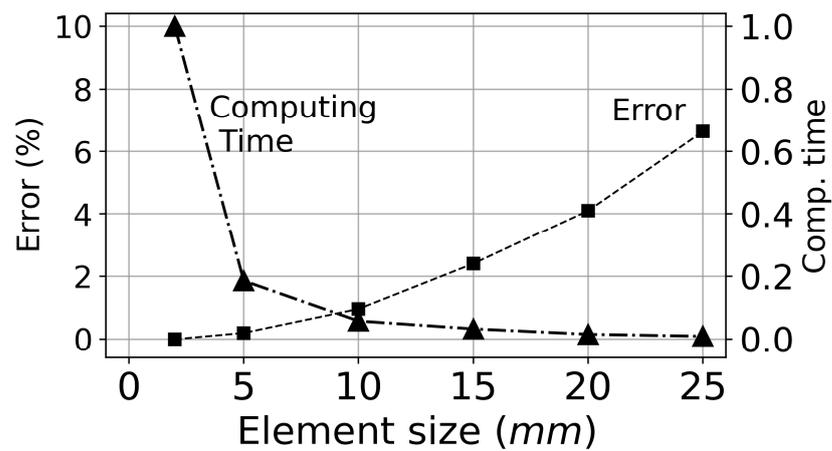


Figure 5. Analysis error and computational time vs. element size.

Adopting a 5 mm element length, an error of 0.2% can be achieved with relatively small computing times. As the overlapping meshed areas corresponding to the plate and the frame have equal element size and nodes at identical positions, they were connected by means of a simple nodal merging operation. Due to this modelling simplification, no friction or slipping effects between plate and frame is considered. During the loading, the plate expansions are constrained by the much stiffer frame, that is, in any case, allowed to expand freely. For all the analysis it was assumed that both plate and boundary conditions are always at the same temperature.

A set of linear buckling analysis for laminates with stacking $[\theta / -\theta]_{2s}$ is displayed in Figure 6, ranging the value of θ from 0° to 90° in 5° increments. A total of five curves, each one collecting the eigenvalues from a particular mode shape, are reported. For each value of θ , at least two eigenvalues are reported: the lowest eigenvalue defines the buckling temperature ΔT_{b1} as well as its corresponding buckling shape; conversely, higher eigenvalues indicate that the plate has alternative states of equilibrium at a temperature higher than ΔT_{b1} . For a plate to naturally jump into another mode shape, the final-state mode shape must be captured as a higher eigenvalue in the linear buckling analysis. For instance, for $\theta = 30^\circ$, the plate reaches buckling at a $\Delta T_{b1} = 57.62^\circ\text{C}$ under mode (1,1), and a second eigenvalue for mode shape (1,2) at $\Delta T = 68.12^\circ\text{C}$.

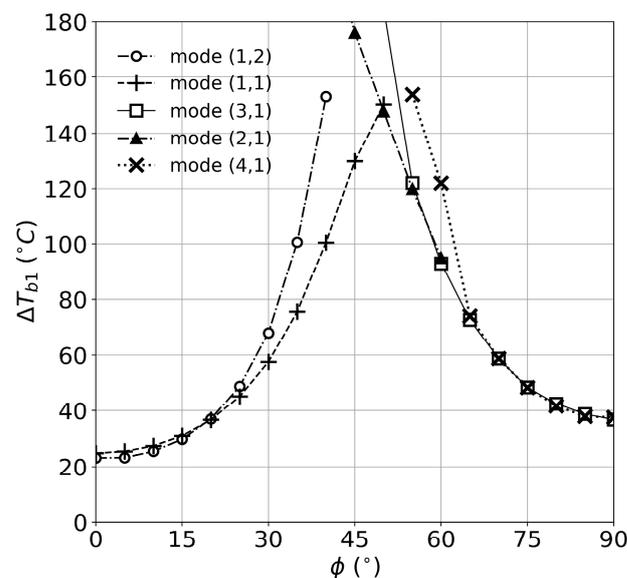


Figure 6. Linear buckling temperatures for $[\theta / -\theta]_{2s}$ laminates.

The results for buckling temperatures obtained in Figure 6 can be divided in intervals: for values of θ within the range $[0^\circ, 20^\circ]$ plates buckle under a (1,2) mode; eigenvalues for modes (1,1) and (1,2) are however very close. Within the range $\theta = [25^\circ, 40^\circ]$ plates buckle under a (1,1) mode shape, and the distance between the curves corresponding to shapes (1,1) and (1,2) increases with angle θ . In this range, mode jumps from a (1,1) to a (1,2) buckling shape are possible. For range $\theta = [45^\circ, 90^\circ]$, mode (1,2) is no longer present. The range of interest for $[\theta / -\theta]_{2s}$ laminates, valid for the selected plate geometry and boundary conditions, is defined by $\theta = [25^\circ, 40^\circ]$.

After having found a range of interest, numerical predictions for the post-buckling behavior of three composite plates with $\theta = 25^\circ, 30^\circ$ and 35° are presented in Figure 7a. For these analyses, an initial imperfection with the shape of the 1st buckling mode and amplitude of 0.5% of plate thickness was assumed. Being the imperfection amplitude small, the nonlinear behavior comes to resemble the ideal plate nonlinear behavior. For the three plates, once ΔT_{b1} is trespassed, out-of-plane deflections start increasing, assuming a (1,1) buckling shape. For $\theta = 25^\circ$ the mode jump is reached at $\Delta T_{b2} = 52.8^\circ\text{C}$, for $\theta = 30^\circ$ this happens at $\Delta T_{b2} = 91.2^\circ\text{C}$, and for $\theta = 35^\circ$ the mode jump occurs at $\Delta T_{b2} = 192.8^\circ\text{C}$ that is beyond the maximal temperature of 120°C .

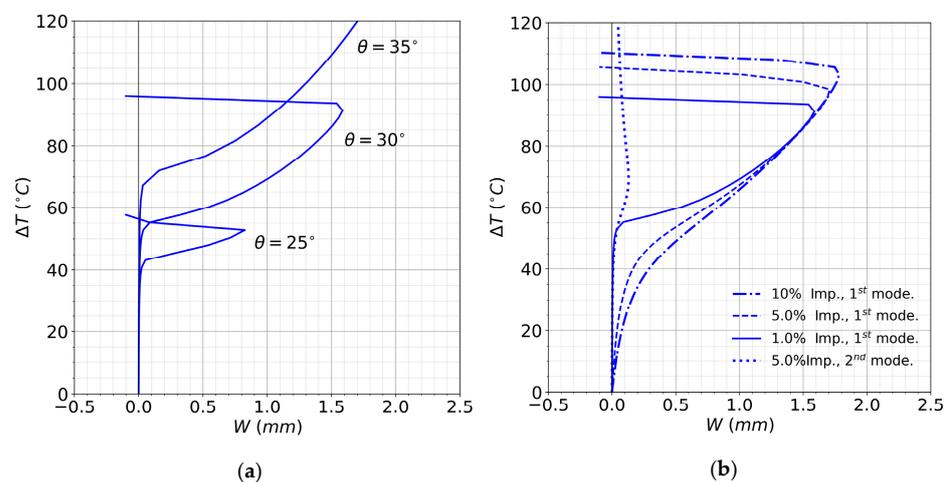


Figure 7. Mode jumping of composite plates: (a) Mode jumping temperature for $[\theta / -\theta]_{2s}$ laminates; (b) Effect of imperfections for $[30 / -30]_{2s}$ laminate.

It can be observed that for the studied range of θ , mode jumping temperature ΔT_{b2} and maximal plate deflections tend to increase with angle θ . This trend is also observed for linear buckling temperature ΔT_{b1} , reported in Figure 6. The reasons are the decrease of α_y and the increase of bending stiffness D_{22} along Y direction for growing values of θ , as reported in Figure 4a,b. It must also be noted that for the analyzed range of θ , the laminate CTE α_x has a negative value, stretching the plate when it is heated and therefore stabilizing the plate against buckling.

The effect of plate imperfections is illustrated in Figure 7b, where the post-buckling behavior of the composite plate with $[30 / -30]_{2s}$ laminate and different imperfections is reported. Three analyses were conducted with an initial imperfection with the shape of a (1,1) buckling mode and amplitudes of 1%, 5% and 10% of plate thickness, plus an analysis with a (1,2) buckling mode imperfection and amplitude of 5% of plate thickness. In particular, ΔT is plotted versus the out-of-plane deflections W , measured in the center of the plate. The plate with the 1% imperfection amplitude, represented as a continuous line, starts deflecting out of plane at a temperature close to the linear eigenvalue buckling temperature reported in Figure 6. Then, the deflection W keeps growing till an abrupt mode change happens at $\Delta T_{b2} = 91.2^\circ\text{C}$. An analogue behavior can be observed in the curves for 5% and 10% imperfection amplitude, reported in Figure 7b as dashed and dot-dashed lines, where ΔT_{b2} tends to increase with imperfection amplitude up to 96.0°C .

and 100.8 °C respectively. However, if an initial imperfection with the shape of the 2nd mode is assumed instead, the mode jump temperature decreases, and with a sufficiently large imperfection amplitude, the plate buckles directly under a (1, 2) mode, skipping the (1, 1) lowest eigenvalue shape.

A parametric study is so conducted for $[\theta / - \theta]_{2s}$ laminates within the range of interest $\theta = [25^\circ, 40^\circ]$ and the results are presented in Figure 8a, where both linear and nonlinear plate behavior are reported. The black bold line collects the 1st eigenvalues or buckling temperatures obtained in the linear buckling analysis, while the black thin line reports the 2nd linear eigenvalue: for instance, considering $\theta = 30^\circ$, the figure shows that the plate has the first eigenvalue at $\Delta T_{b1} = 57.62$ °C, and has a second eigenvalue at $\Delta T = 68.12$ °C with a mode shape (1, 2). The blue curves report the mode jumping temperatures for different imperfection shapes and amplitudes: the blue bold line collects values of ΔT_{b2} for plates with 0.5% 1st mode imperfection amplitude, the blue dot-dashed and dotted lines report values of ΔT_{b2} for 5% and 10% imperfection amplitude, and at last, a blue dashed line captures values of ΔT_{b2} for a 5% imperfection shape with a 2nd mode shape. Considering a plate with $\theta = 30^\circ$ and 0.5%, 5% and 10% of 1st mode imperfection amplitudes, the plate would find a mode jump at $\Delta T_{b2} = 91.2$ °C, 96.0 °C and 100.8 °C respectively, matching values previously reported in Figure 7b. It can be observed how for all analyzed values of θ , increasing amplitudes of 1st mode shape imperfection tend to rise ΔT_{b2} . Conversely, 2nd mode imperfections tend to decrease ΔT_{b2} . Within the range $\theta = [25^\circ, 30^\circ]$, the lines for ΔT_{b1} and ΔT_{b2} overlap, meaning that the plate buckles directly under a (1, 2) shape. $[35 / - 35]_{2s}$ layup shows a ΔT_{b2} beyond 120 °C, so the remaining candidate layups are $[25 / - 25]_{2s}$ and $[30 / - 30]_{2s}$.

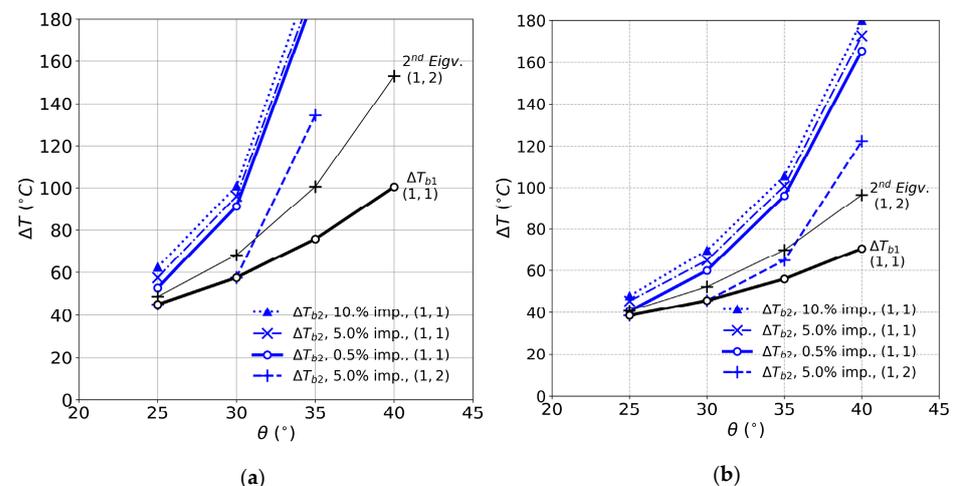


Figure 8. Mode jumping temperature for different laminates: (a) $[\theta / - \theta]_{2s}$ laminates; (b) $[\theta / - \theta / \varphi / - \varphi]_s$ laminates with $\varphi = \theta - 25$.

In the third and last step, the stacking orientations were improved by fine-tuning the layups. The objective was to make them more suitable for experimentation by decreasing both ΔT_{b1} and ΔT_{b2} . This can be achieved by using a generic angle-ply laminate in the form $[\theta / - \theta / \varphi / - \varphi]_s$, where both angles are smaller than 45° and the outer layer angle θ is significantly larger than φ . Taking for example $\varphi = \theta - 25^\circ$, the bending stiffness along Y axis D_{22} does not decrease substantially, as D_{22} is mainly formed by the contributions of the outer plies and angle θ remains constant. On the other hand, α_y increases significantly, and also α_x increases, changes its sign and becomes positive. Results for $\varphi = \theta - 25^\circ$ are reported in Figure 8b for the range $\theta = [25^\circ, 40^\circ]$. For the considered values of θ , it can be appreciated a decrease of both ΔT_{b1} and ΔT_{b2} . As a consequence, laminates with $\theta = 35^\circ$ show a mode jump within 120 °C range, even if a moderate 2nd mode imperfection is present. Consequently, two stacking orientations of the type $[\theta / - \theta / \varphi / - \varphi]_s$ are selected for experimentation: $[30 / - 30 / 5 / - 5]_s$ and $[35 / - 35 / 10 / - 10]_s$.

4. Test Setup

Three specimens were manufactured and tested. The first plate was made in aluminum alloy 2024T3 [31] with a thickness of 1 mm. The aim of this first test was assessing the functionality of the test setup and providing a benchmark using an isotropic material with well-defined properties. The second and third specimens were made of AS4/8552 composite material, with stacking orientations $[30/-30/5/-5]_s$ and $[35/-35/10/-10]_s$. For the manufacturing of the plates, each individual layer was cut in an automated cutting table in order to ensure the accuracy in the ply orientation. After that, the layers were stacked by hand and ultimately cured in an autoclave.

For the tests, the plate is fixed in the frame, which has the function of providing support to the plate as well as restraining its thermal expansion during the test, as shown in Figure 8. The frame is composed of two symmetric parts between which the plate is placed and then assembled together with steel bolts. The frame is made of Invar 36 material, which has a very low CTE.

Before a test starts, the plate and the frame have to be bolted together. The frame relies on the bolted joint to restraint the thermal expansion in the plate. The selected steel bolts have a CTE of $12.2 \mu\text{m}$, which is much higher than the CTE of Invar 36 material. The assembly torque needs to be high enough to ensure that not too much bolt pretension is lost through heating due to the difference in CTE between frame and bolts. A total assembly torque of 60 Nm is gradually brought to the fixture in three levels, following a crossed path to ensure uniform compression throughout the frame.

To induce the temperature increment, a Vötsch brand oven was used (Vötsch Industrietechnik GmbH, Reiskirchen, DE). It had a volume capacity of $600 \text{ mm} \times 600 \text{ mm} \times 600 \text{ mm}$. A metallic steel support, reported in Figure 9, was used to keep the frame in a vertical position.

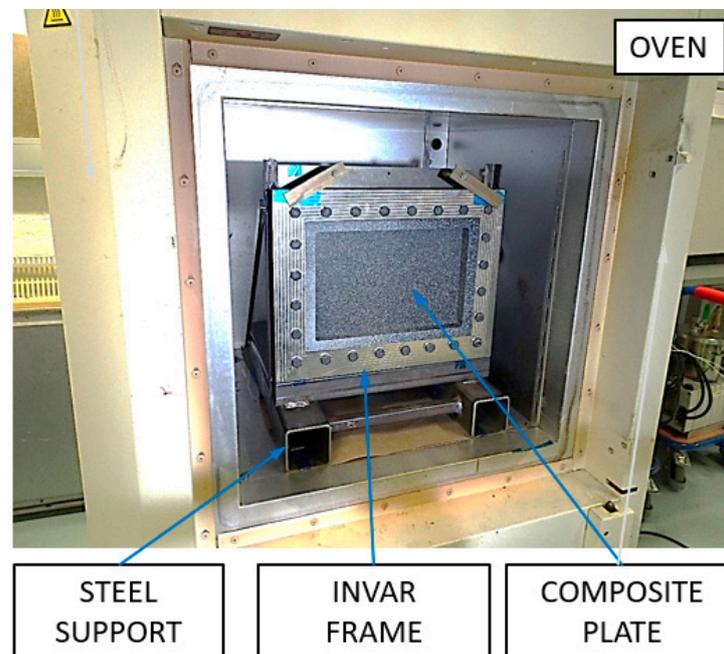


Figure 9. Specimen inside oven.

The test setup is presented in Figure 10. For the test, the original door of the oven is removed and substituted by a replacement wooden hatch with an inner ceramic lining. This hatch has a rectangular hole that allows direct visual contact with the plate for measurement acquisition. A rectangular, metallic adaptor with the same cross section as the hole is used for sealing gap between frame and hatch.

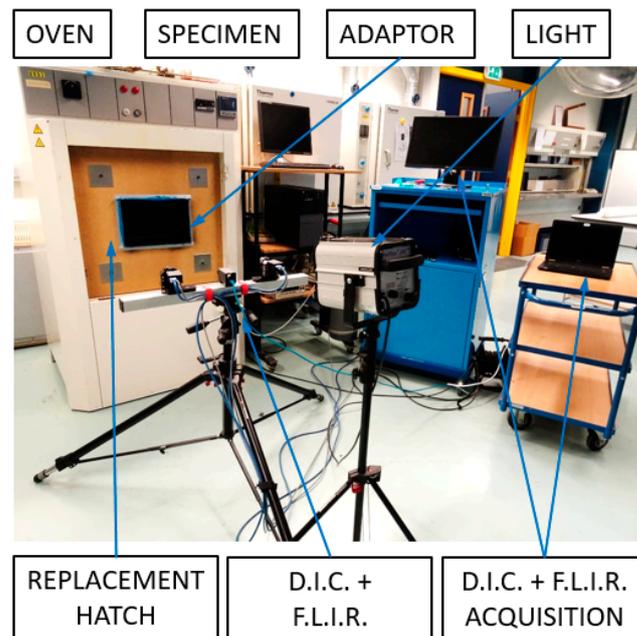


Figure 10. Test setup.

The acquisition systems collect measurements for both displacement and temperature of the plate during the complete duration of the experiment. The measurement of the out-of-plane displacement is performed with a Vic3D Digital Image Correlation (DIC) system (Correlated Solutions, Irmo, SC, USA), equipped with two cameras. The plate has to be painted with a white speckle pattern over black background. During the tests, the temperature is monitored using a thermocouple sensor placed at the center of the plate and using an A65 thermocamera (FLIR Systems, Arlington, VA, USA) placed on the outside of the oven.

In the FE analysis the plate is loaded by applying temperature increments with respect to the initial reference temperature. In order to being able to compare experimental results with numerical analysis, a reference temperature T_{ref} must be added to applied ΔT in the analysis. Indeed it is considered that at T_{ref} the plate is in an in-plane stress free state. T_{ref} was calculated here as the average temperature before and after bolting operation, measured in the fixture. Thus, the equation $T_{analysis} = T_{ref} + \Delta T$ holds.

The tests consist in the gradual, monotonic heating of the plate and frame. Thermocouples start recording 10 minutes before test start. Once the test initiates, DIC and FLIR thermocamera system start recording data. For all measurement system, an acquisition rate of 1 measurement every 10 s was used.

5. Thermal Tests

The first test was performed with the plate made in aluminum alloy 2024T3 of 1 mm thickness. The temperature path measured by the thermocouple is displayed in Figure 11. The goal temperature path into the oven is represented as a black bold line, while temperature measured in the thermocouple is represented as a red bold line. A large difference between the two curves can be appreciated. It is believed this difference was caused by air convection on the external face of the plate.

The contour plots for numerical and experimental deflections are presented in Figure 12. For all numerical analyses, a 1st mode shape imperfection with a 10% amplitude was considered. The obtained experimental plots show reasonably good correlation with the numerical results: predicted and obtained buckling shapes are matching in shape, even if the experimental values are larger than the predictions.

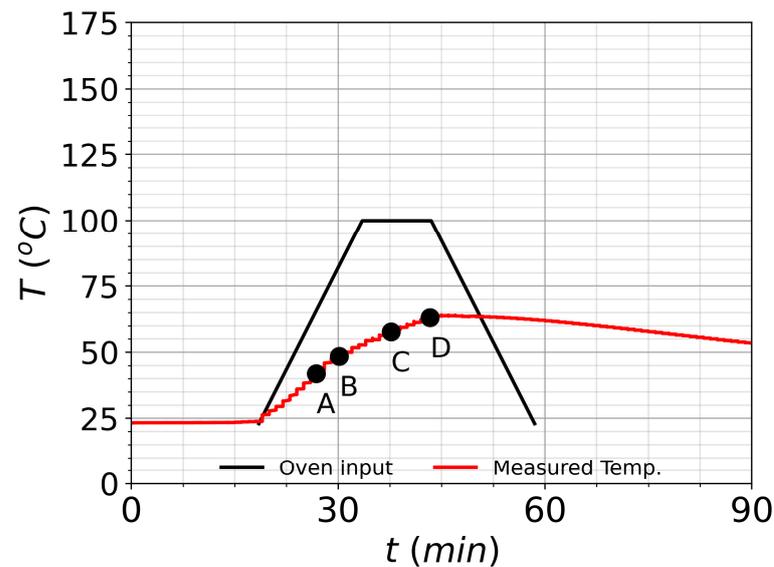


Figure 11. Heating cycle for aluminum alloy plate.

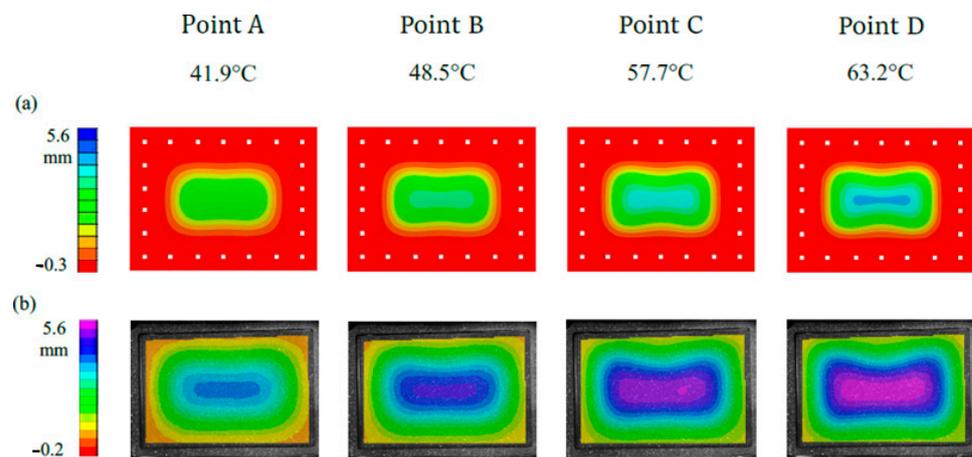


Figure 12. Out-of-plane deformation for aluminum alloy plate: (a) FE analysis; (b) Test.

This becomes evident in the curves plotted in Figure 13 where Z-coordinate was plotted versus the temperature for the three plates. For the aluminum alloy plate the Z-coordinate was measured at plate position $(a/2, b/2)$. Two curves are plotted, one blue dotted and a black curve with round markers representing the FE and experimental results, respectively. Even if the experimental deflections result to be noticeably larger, the numerical prediction captures the essence of the plate nonlinear behavior. The metallic plate shows no mode jumping.

The second test was performed on the composite plate with $[30/-30/5/-5]_s$ layup. A graph of temperature as a function of time is reported in Figure 14. The ideal temperature cycle is represented as a black bold line, labelled as “Oven input”. In practice the plate never reaches the programmed temperature, so a prediction was performed to ensure that the desired temperature was reached; this estimation, represented as a bold violet line, was performed using data from two preliminary tests, where the plate was heated at two different temperature rates. This violet line was also used for the numerical simulation of the plate. Using the differences between programmed and obtained temperature rates in both experiments, for a desired plate temperature rate an oven input program rate can be extrapolated.

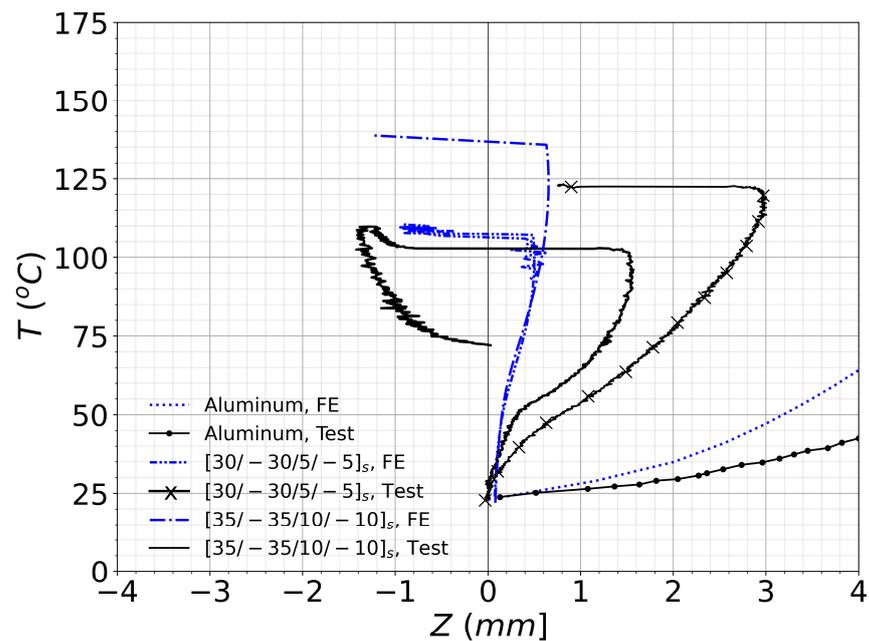


Figure 13. Temperature vs. out-of-plane deformation for the three plates.

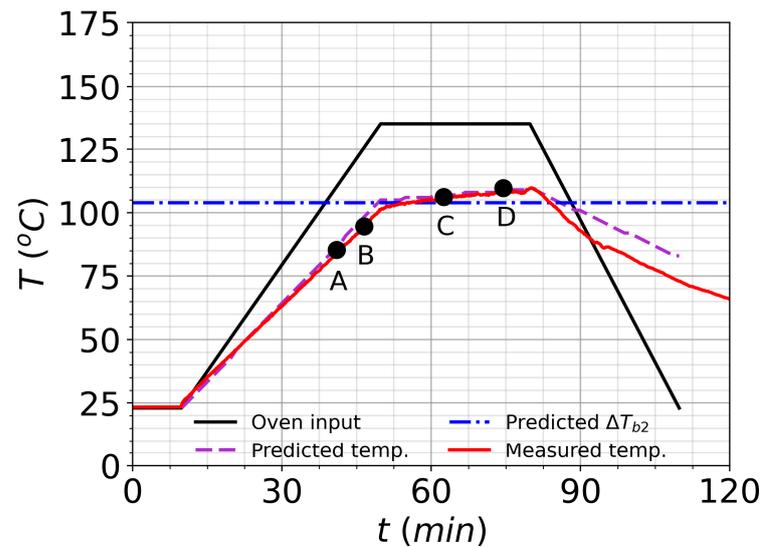


Figure 14. Heating cycle for $[30/-30/5/-5]_s$ composite plate.

The expected value for mode jumping is plotted in the graph as a horizontal dot-dashed blue line. The temperature measured in the plate with the thermocouple is depicted as a red bold line. It closely matches the predicted temperature curve, in violet, and crosses it at the numerically expected temperature ΔT_{b2} . In Figure 15 experimental and numerical plots are reported. Temperature points A and B correspond to the points in which the plate is heated and the out-of-plane displacements are clearly visible in the form of a (1,1) buckling shape. At temperature values C and D a change in mode buckling into a (1,2) shape is registered. When observing the FE simulations at point C the mode jump has still not happened.

This difference becomes evident when observing Figure 13, where plate Z-position is measured at $(a/2, b/4)$. In the two curves for laminate $[30/-30/5/-5]_s$, the black line for the test and the dot-dashed blue line for the FE analysis are displayed. A change on sign in Z coordinate takes place at 107°C for the numerical prediction and at 102.5°C for the experimental result. Thus, the mode jump took place slightly earlier than what the FE simulation indicates. Additionally, 7.5 minutes after point D the temperature in the plate

becomes maximum and after that it starts decreasing. When the plate cools down to the mode jump temperature $\Delta T_{b2} = 102.5^\circ\text{C}$, it does not spontaneously jump back into the original one half-wave configuration but it remains at a (1,2) shape till $T = 72^\circ\text{C}$ showing some sort of hysteresis behavior.

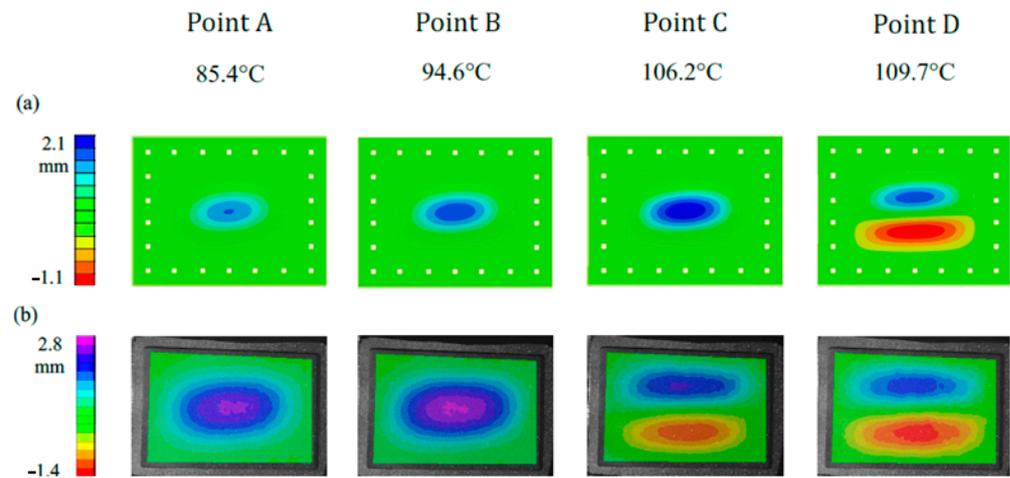


Figure 15. Out-of-plane deformation for $[30/ - 30/5/ - 5]_s$ composite plate: (a) FE analysis; (b) Test.

The third test was performed on the composite plate with $[35/ - 35/10/ - 10]_s$ laminate. The heating diagram in Figure 16 shows that the curves for experimental temperature and estimated mode jump temperature do not cross, hence the temperature estimation was not accurate enough. However, in Figure 17, the experimental image of point D shows that a mode jump is indeed captured, differing from FE prediction which still shows a (1,1) buckling mode. This mode jump is also observable in Figure 13, where, although the sign of Z does not change, it experiences a sudden drop at 122.5°C . As occurred with previous plate $[30/ - 30/5/ - 5]_s$, the predicted mode jump temperature happens at a larger temperature, at 137°C .

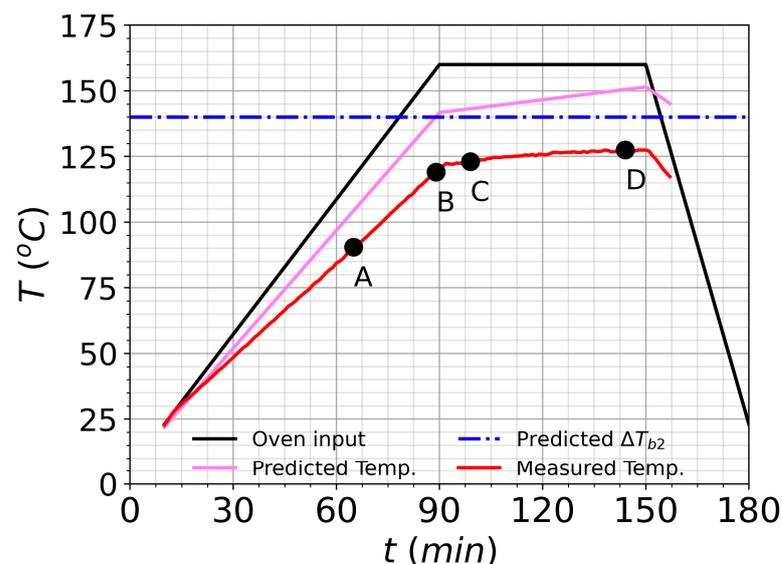


Figure 16. Heating cycle for $[35/ - 35/10/ - 10]_s$ composite plate.

A brief comparison of the three plates is here presented. The curves in Figure 13 indicate that for all cases plates show larger deflections than in the FE predictions: they are 33% higher for the $[30/ - 30/5/ - 5]_s$ plate and 68% for the $[35/ - 35/10/ - 10]_s$ plate. In the case of the two tested composite plates, the plate with layup $[35/ - 35/10/ - 10]_s$

shows much larger deflections than the plate with $[30/-30/5/-5]_s$ layup. Both plates present a mode jump, being the temperature ΔT_{b2} higher for the $[35/-35/10/-10]_s$ plate. For both composite plates the mode jump manifests at a slightly lower temperature than in the numerical prediction.

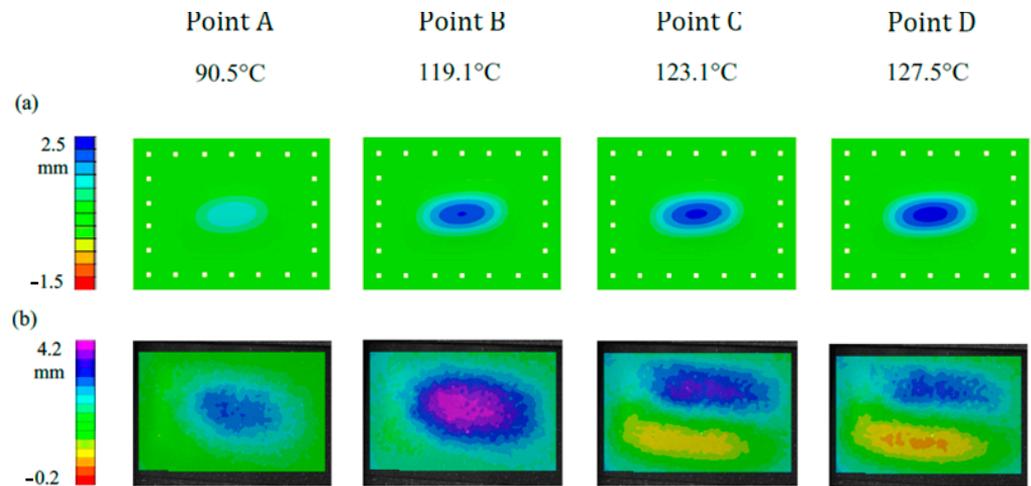


Figure 17. Out-of-plane deformation for $[35/-35/10/-10]_s$ composite plate: (a) FE analysis; (b) Test.

The discrepancies found between predictions and experimental results can be attributed to multiple factors. In the FE models, possible source of uncertainty is mainly the temperature dependency of material properties. Regarding the tests, the sources can be different: deviation from nominal values in material properties, change in material properties due to sustained heating, small variations in the ply angles due to the manufacturing process, relaxation in the boundary conditions due to the thermal expansion in the steel bolts, irregular heating in the specimen and geometrical imperfections of the plate. Even though the differences with the prediction are evident, the dynamic explicit procedure successfully captures the tendency to show mode jumping in the studied composite plates when these are under heating loading condition. Numerical predictions can be improved by individually addressing all previously mentioned factors. However, even though these deserve independent attention, they bring complexity and they therefore remain out of scope for the present investigation. Only the initial imperfection over ΔT_{b2} is addressed here: as shown in Section 3, 1st mode imperfection patterns with a 5% amplitude tend to stabilize the nonlinear range of the plate rising ΔT_{b2} , while plate imperfections dominated by other modes shapes may decrease the mode jumping temperature or even neutralize the effect completely. All FE predictions reported in Figure 13 used a 1st mode, 10% imperfection amplitude for all cases, while more adverse imperfections may have been present. Indeed, buckling fixtures can imprint a certain imperfection in the plate which can influence its buckling shape. Finally, the position of the heating source is also noteworthy: the tested plate is heated only from one side, and due to this, a through-thickness temperature gradient appears in the plate. This creates a thermal moment that can help the plate overcoming potentially disruptive initial imperfections, buckling into a (1,1) mode shape, as all three experiments buckle under a (1,1) shape, with the maximum deflection area directed towards the interior of the oven.

Based on previous results, 1st mode geometrical imperfections and slight panel curvatures could be used deliberately to stabilize the post-buckling shape in composite panels and help better control the mode jumping effect. Mild geometrical patterns such as panel beadings have been successfully used in the past for structural stabilization of thermally post-buckled structures [32], and may offer interesting properties towards applications of mode jumping. Also, morphing applications using mode jumping should take advantage of heating asymmetries present in aerospace applications such high-speed aircraft or

satellites, where the structure is heated in only one side due to aerodynamic friction or solar radiation.

6. Conclusions

In this study the thermal buckling and post-buckling behavior of composite plates was investigated. Plates were studied using both numerical and experimental techniques. A methodology for selecting composite plates showing mode jumping under heating was presented. Linear buckling eigenvalue analysis proved useful for the identification of desired secondary mode shapes, while dynamic explicit analysis was able to capture this event and give estimations for mode jumping temperatures. Additionally, parametric analysis predicted a stabilizing influence of 1st mode imperfections with diverse amplitudes over the post-buckling shape increasing the mode jumping temperature, while 2nd mode imperfections of comparable amplitude appeared to have a destructive effect over it. Thus, inducing (1,1) imperfections on composite panels could be advantageous if mode jumps are sought. As a result of the preliminary analysis, two plates made in AS4/8552 composite material with layups $[30/ - 30/5/ - 5]_s$ and $[35/ - 35/10/ - 10]_s$ were selected for experimentation in a novel test setup. An additional plate made in 2024T3 aluminum alloy was selected as a benchmark case and to demonstrate the functionality of the setup. The post-buckling behavior was successfully reproduced in all plates, while mode jumping was successfully captured for both composite plates. The experimental deflections resulted larger than the predicted values for all three tested plates. For the composite plates, mode jumping happened at temperatures lower than predicted, being the largest difference limited to 13%. Even if discrepancies with numerical predictions are reported, nonlinear numerical procedures succeeded in capturing the trends and overall behavior in all plates. It is believed that sources for discrepancies may range from temperature dependent material properties, manufacturing or assembly defects. Also, potential morphing applications using mode jumping should take advantage of heating asymmetries present in aerospace applications. These results can be valuable as a preliminary study when considering applications of thermally triggered buckling shape change in aerospace applications.

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